

1.2 Do problem 1.1 with $M = \alpha T$ where $\alpha = 0$ to ∞ .

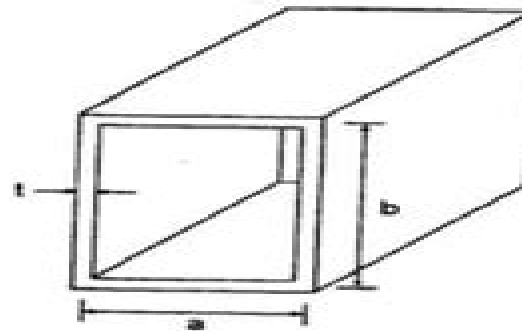


Figure 1.16 Closed thin-walled section

Solution:

(1) The bending stress of beams is $\sigma = \frac{My}{I}$, where y is the distance from the neutral axis. The moment of inertia I of the cross-section can be calculated by considering the four segments of thin walls and using the formula for a rectangular section with height h and width w , $I = \sum (\frac{I}{12} wh^3 + Ad^2)$ in which A is the cross-sectional area of the segment and d is the distance of the centroid of the segment to the neutral axis. Note that the Parallel Axis Theorem is applied. The result is $I = 2 \cdot \frac{1}{12} tb^3 + 2 \cdot [\frac{1}{12} \cdot at^3 + (at) \cdot (\frac{b}{2})^2] \approx \frac{tb^2}{6} (3a + b)$, assuming that t is very small.

(2) The shear stress due to torsion for a closed thin-walled section shown above is

$$\tau = \frac{T}{2abt}.$$

(3) Two approaches are employed to find the solution.

(i) Assume that the bending stress reaches the allowable $\sigma_{allowable}$ first and find the corresponding bending maximum bending moment. Then apply the stated loading condition of $M = \alpha T$ to check whether the corresponding τ_{max} has exceeded the allowable shear stress $\tau_{allowable}$. If this condition is violated, then the optimized b/a ratio is not valid.

$$(a) \quad \sigma \Big|_{r=\frac{b}{2}} = \frac{My}{I} = \frac{M \cdot \frac{b}{2}}{\frac{tb^2}{6} (3a + b)} = \frac{3M}{tb(3a + b)}$$

When given $I = 2(a + b)$ as a constant, a can be expressed in terms of b

Mechanics Of Aircraft Structures Solutions

Bruce K. Donaldson



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Scientific and Technical Aerospace Reports ,1995 Lists citations with abstracts for aerospace related reports obtained from world wide sources and announces documents that have recently been entered into the NASA Scientific and Technical Information Database

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Government Research Reports, 1964 **Damage tolerance in aircraft structures**, 1971 **Virtual Principles in Aircraft Structures** M. Gatewood, 2012-12-06 The basic partial differential equations for the stresses and displacements in classical three dimensional elasticity theory can be set up in three ways 1 to solve for the displacements first and then the stresses 2 to solve for the stresses first and then the displacements and 3 to solve for both stresses and displacements simultaneously These three methods are identified in the literature as 1 the displacement method 2 the stress or force method and 3 the combined or mixed method Closed form solutions of the partial differential equations with their complicated boundary conditions for any of these three methods have been obtained only in special cases In order to obtain solutions various special methods have been developed to determine the stresses and displacements in structures The equations have been reduced to two and one dimensional forms for plates beams and trusses By neglecting the local effects at the edges and ends satisfactory solutions can be obtained for many case The procedures for reducing the three dimensional equations to two and one dimensional equations are described in Chapter 1 Volume 1 where the various approximations are pointed out **Shock and Vibration, Data Analysis and Applications** Defense Documentation Center (U.S.), 1963 *Technical Abstract Bulletin*, 1964 *Government-wide Index to Federal Research & Development Reports*, 1966 **Technical Data Digest**, 1951 **Technological Forecasting for Decisionmaking** Joseph Paul Martino, 1972

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